

WE CLAIM

1. A wall element for a wall structure of a gas turbine engine combustor, the wall element including at least one
5 surface, the surface, in use, faces in a downstream direction relative to the general direction of fluid flow through the combustor, wherein said downstream facing surface comprises a thermally resistant material.
2. A wall element according to claim 1 including a main
10 body member, the main body member comprising upstream and downstream edges, wherein the downstream edge of said main member has a downstream facing surface, the downstream facing surface comprising said thermally resistant material.
3. A wall element according to claim 1 wherein a plurality
15 of heat removal members are provided on the main body member, each of said heat removal member furthest downstream including a downstream facing surface, the downstream facing surface comprising thermally resistant material.
4. A wall element according to claim 3 wherein the wall
20 element comprises a tile.
5. A wall element according to claim 3 wherein the heat removal members are in the form of pedestals.
6. A wall element according to claim 3 wherein said
thermally resistant material extends substantially the whole
25 length of said heat removal members.
7. A wall element according to claim 3 wherein the heat removal members are upstanding from the main body member.
8. A wall element according to claim 7 wherein the heat removal members have a substantially circular cross-section.
- 30 9. A wall element according to claim 7 wherein the thermally resistant material is provided on a downstream facing arc of said surface.
10. A wall element according to claim 9 wherein said arc subtends an angle of at least substantially 90° of said
35 surface.
11. A wall element according to claim 9 wherein the arc

subtends an angle of at least substantially 180° .

12. A wall element according to claim 9 wherein the arc subtends an angle of no more than substantially 180° .

13. A wall element according to claim 1 wherein the
5 thermally resistant material is a thermal barrier coating.

14. A wall element according to claim 13 wherein the thermal barrier coating is magnesium zirconate.

15. A wall element according to claim 13 wherein the thermal barrier coating is yttria stabilised zirconia.

10 16. A combustor for a gas turbine engine having a wall structure comprising inner and outer walls, wherein the inner wall comprises a plurality of wall elements as claimed in claim 1.

17. A gas turbine engine incorporating a combustor as
15 claimed in claim 16.